

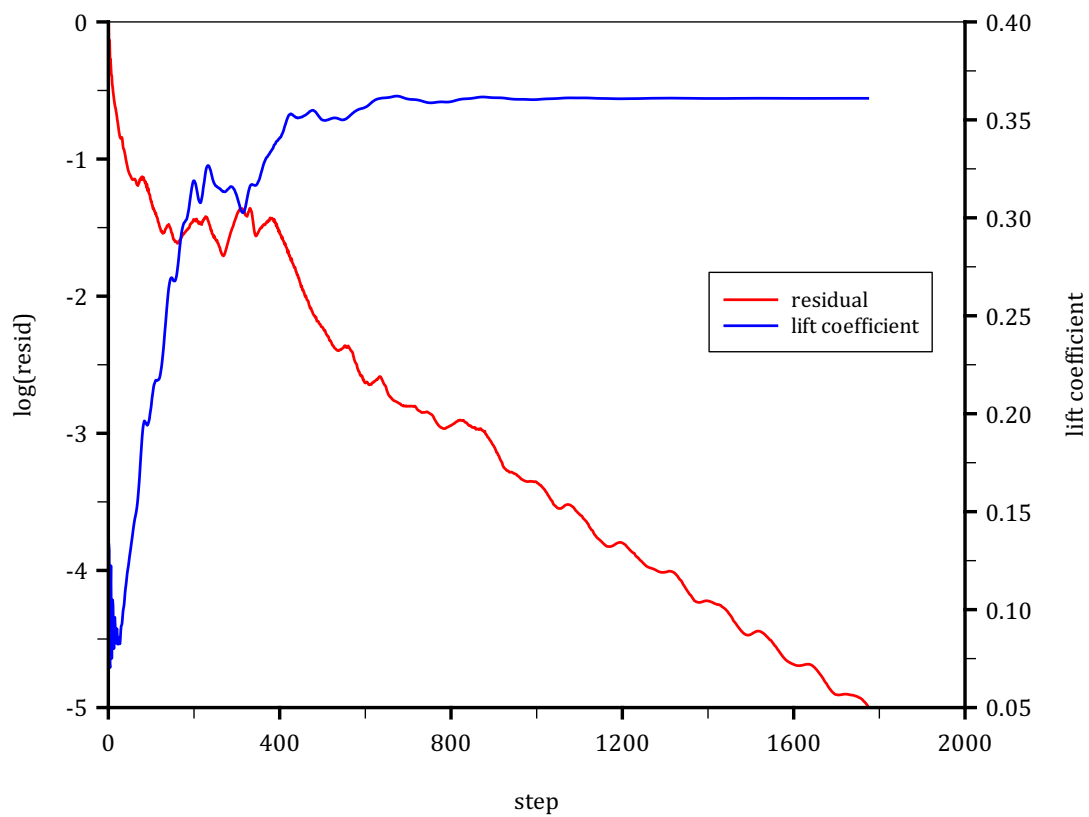
Solution of 2-D Euler Equations: NACA 0012 Airfoil

Spatial discretization by Roe's upwind scheme:

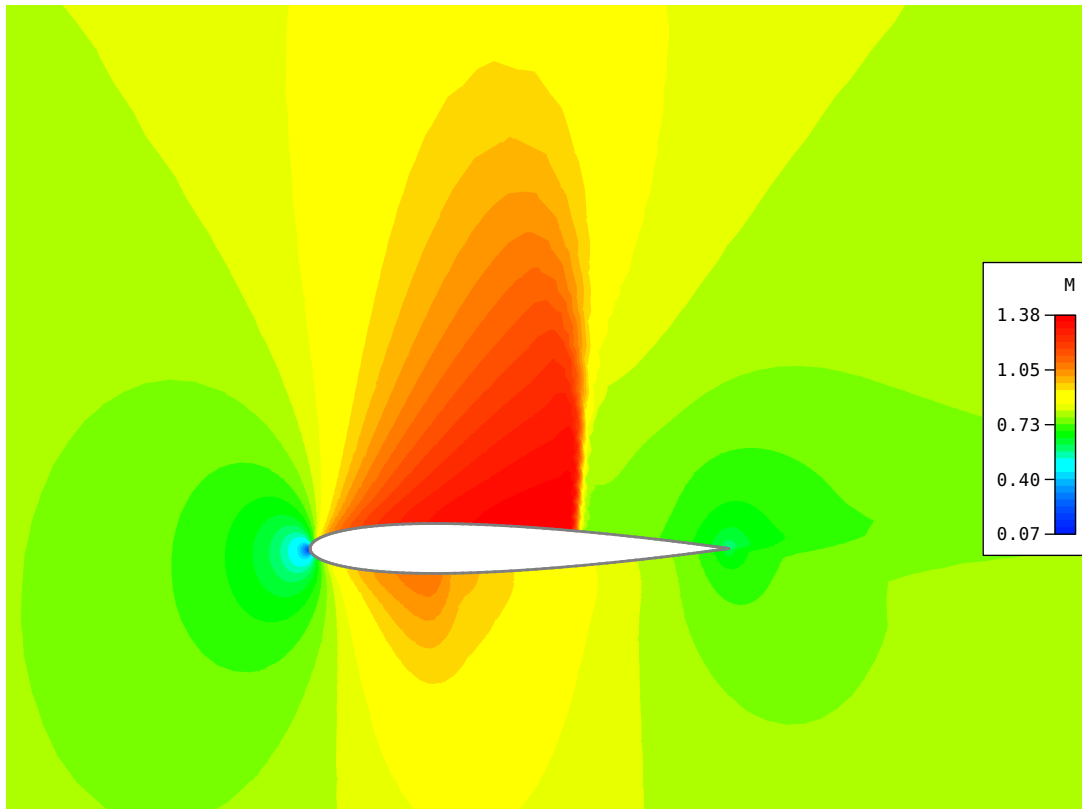
$$\sigma = 5.5, \varepsilon = 0.4, K = 5$$

Boundary conditions:

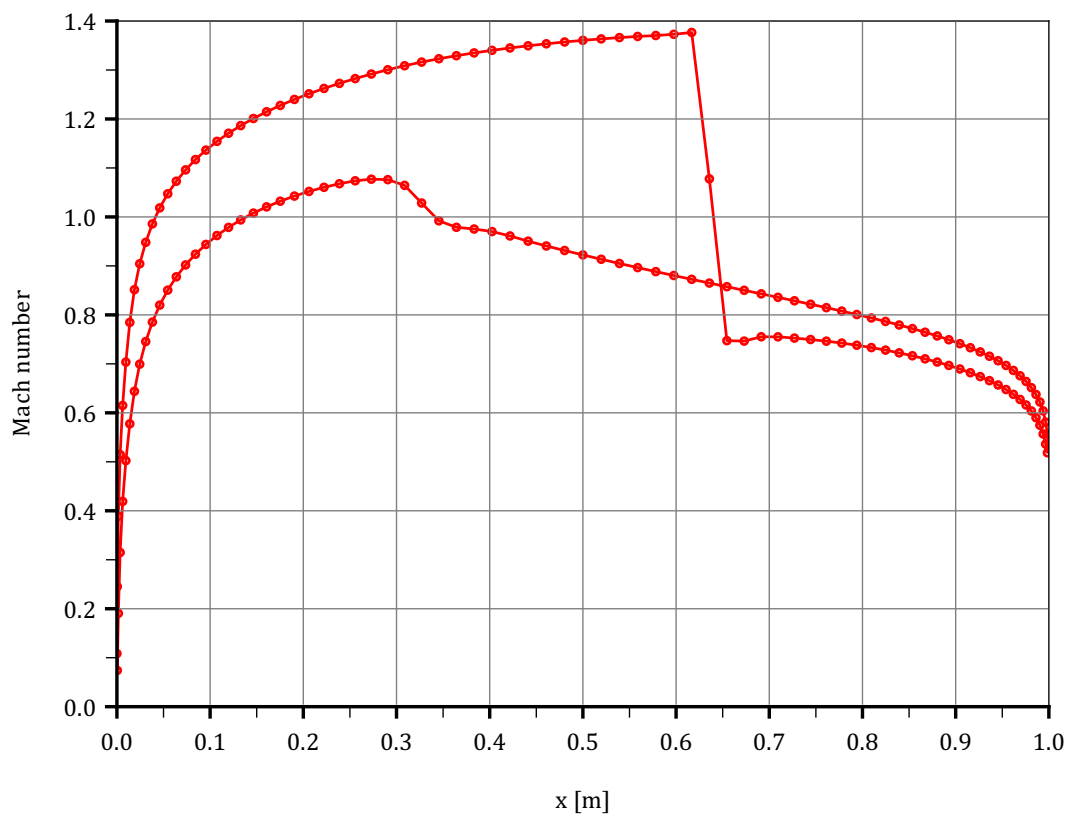
$$M_\infty = 0.8, \alpha = 1.25^\circ, p_\infty = 1.0 \cdot 10^5 \text{ Pa}, T_\infty = 288.0 \text{ K}.$$



Convergence history.



Mach number distribution around the airfoil.



Mach number over the chord length.